

# LEC: Grav 2-body III

4 March

- constructing orbits from  $r(\theta)$
- "scattering kinematics" (UNBOUND ORBITS)
- Gravity assist

start w/ Page \*

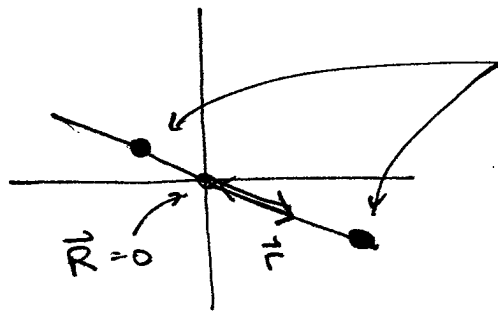
Center of mass frame :  $\vec{R} \dot{=} 0$

choose coordinates s.t.  $\vec{R} = 0$

$$\vec{r}_1 = \frac{m_2}{m_1 + m_2} \vec{r}$$

$$\vec{r}_2 = \frac{-m_1}{m_1 + m_2} \vec{r}$$

Positions of each body are parallel to  $\vec{r}$



which one is closer?  
more massive one  
closer to origin

maybe start w/ "escape"

Simple example : binary stars w/  $m_1 = m_2$

$$\vec{r}_1 = -\vec{r}_2 = \frac{1}{2} \vec{r}$$

ECCENTRICITY : ~~eccentricity~~

$$\epsilon^2 = \left( E^2 \frac{2Q^2}{4K^2} - 1 \right)$$

CONSTANTS

$$V = -k/r$$

0 ≤ ε < 1  
↑  
CIRCLE

hyperbola  
↓

Suppose  $e = \sqrt{3}/2$

recall:  $\frac{a}{b} = \frac{1}{\sqrt{1-e^2}} = \frac{2}{1}$

major's  
minor axes  
of ellipse

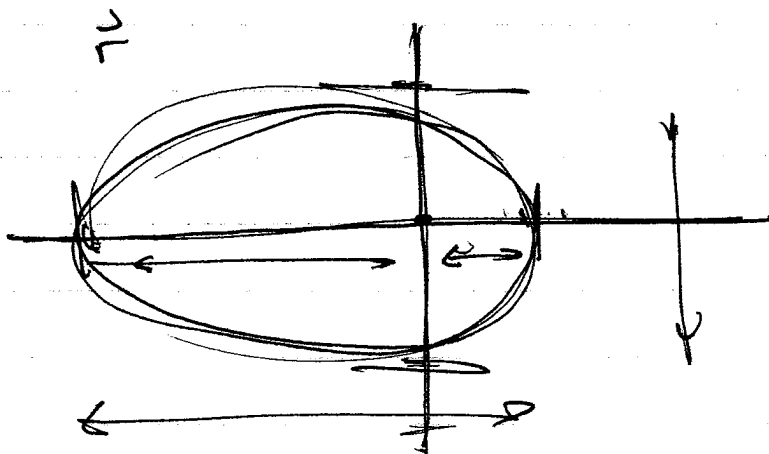
Aspect ratio

so let's draw the orbit

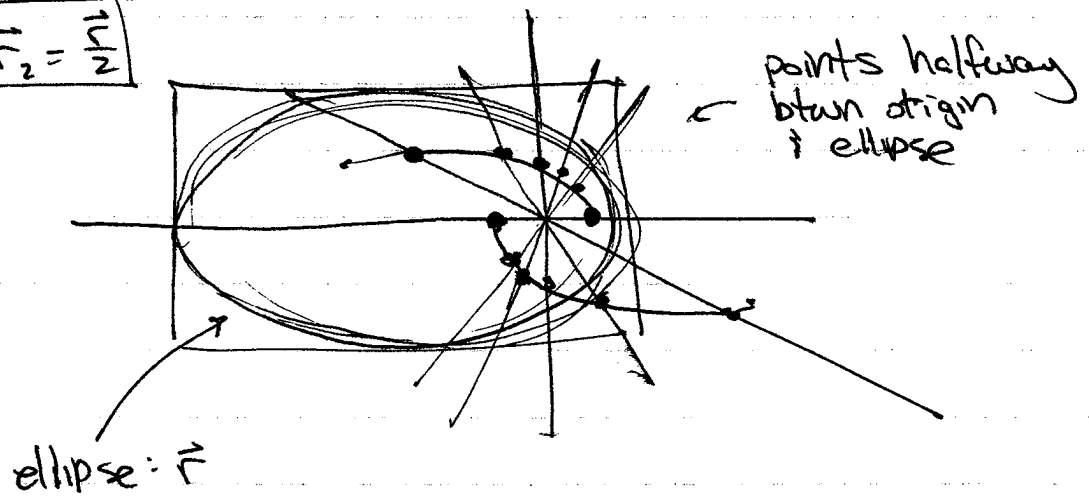
$$r_{\max} = \frac{p}{1+e} \quad \triangleright \quad p = \frac{l^2}{\mu k}$$
$$r_{\min} = \frac{p}{1-e}$$

the origin  $\vec{R} = 0$  divides the major axis according to the ratio

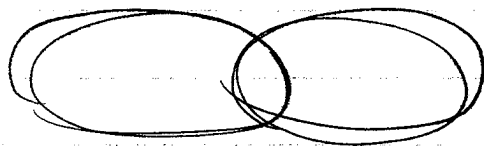
$$\frac{r_{\max}}{r_{\min}} = \frac{1+e}{1-e} \approx 14$$



$$\vec{r}_1 = -\vec{r}_2 = \frac{\vec{r}}{2}$$



If you took a time lapse photo, the binary star system would look like:



Remark: you'll do this in problem #4!

Next: "orbits" with  $\epsilon > 1$

$$\star$$

$$k = GM_1 M_2$$

↓

$$\Leftrightarrow \text{Energy} = (\epsilon^2 - 1) \frac{A}{2r_0} = (\epsilon^2 - 1) \frac{Mk^2}{2l^2}$$

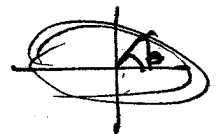
$> 0 \rightarrow$  "ESCAPE TO INFINITY"

by the way: what does this mean?  
TOTAL ENERGY is only defined up to an overall constant, no?

$\hookrightarrow$  w/rt FREE PARTICLE  
in GR, constants matter  
 $\hookrightarrow$  cosmological constant  
(supersymmetry: ~~easy~~ vacuum)

Recall: ellipse

$$r_0 = r + \epsilon r \frac{\cos \theta}{x}$$



$$\rightarrow (r_0 - \epsilon x)^2 = r^2 = x^2 + y^2$$

$$\rightarrow r_0^2 = \left( (1 - \epsilon^2)x^2 + 2\epsilon r_0 x + (1 - \epsilon^2)x_0^2 \right) + y^2$$

$$r_0 \left( 1 + \frac{\epsilon^2}{1 - \epsilon^2} \right) = \frac{r_0^2}{1 - \epsilon^2}$$

$$x_0 = \frac{-\epsilon}{1 - \epsilon^2} r_0$$

$$= (1 - \epsilon^2)(x - x_0)^2 + y^2$$

$$\frac{r_0^2}{1-\varepsilon^2} = (1-\varepsilon^2)(x-x_0)^2 + y^2$$

$$\rightarrow 1 = \underbrace{\frac{(1-\varepsilon^2)^2}{r_0^2}}_{a^2} (x-x_0)^2 + \underbrace{\frac{1-\varepsilon^2}{r_0^2}}_{b^2} y^2$$

$$\rightarrow \boxed{a = \frac{r_0}{1-\varepsilon^2} \quad b = \frac{r_0}{\sqrt{1-\varepsilon^2}}}$$

For ~~ELLIPSE~~:  $\varepsilon < 1$   
HYPERBOLA

$$-\frac{r_0^2}{\varepsilon^2-1} = -(\varepsilon^2-1)(x-x_0)^2 + y^2$$

$$\rightarrow 1 = \underbrace{\frac{(\varepsilon^2-1)^2}{r_0^2}}_{a^2} (x-x_0)^2 - \underbrace{\frac{\varepsilon^2-1}{r_0^2}}_{b^2} y^2$$

$$\boxed{a = \frac{r_0}{\varepsilon^2-1} \quad b = \frac{r_0}{\sqrt{\varepsilon^2-1}}}$$

$$\boxed{1 = \frac{(x-x_0)^2}{a^2} - \frac{y^2}{b^2}}$$

hyperbola



What is this impact parameter?

considers asymptotic expressions for  $l \gg E$ .

$$l = b \mu v_{\infty} = (\vec{r} \times \vec{p})_z$$

$$E = \frac{1}{2} \mu v_{\infty}^2$$



$$v_{\infty} = |\dot{\vec{r}}| \quad \text{in lim } |\vec{r}| \rightarrow \infty$$

$$\frac{l^2}{E} = 2\mu b^2 \Rightarrow \boxed{b^2 = \frac{l^2}{2\mu E}} = \frac{A \mu r_0}{\frac{A \mu}{r_0} (\epsilon^2 - 1)} = \boxed{\frac{r_0^2}{\epsilon^2 - 1}}$$

recall:  $E = (\epsilon^2 - 1) \frac{A}{2r_0} = (\epsilon^2 - 1) \frac{\mu \epsilon^2}{2\epsilon^2}$

IMPACT PARAM.

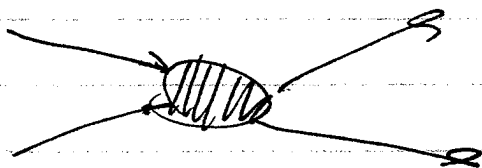
compare to hyperbola equation - in fact,

b is the "b" in hyperbola!

Scattering: one state of uniform (unacc.) motion

↓  
another state →

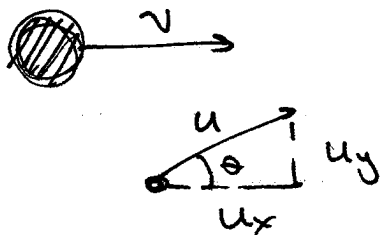
"asymptotic states"



asympt free,  
non-interacting

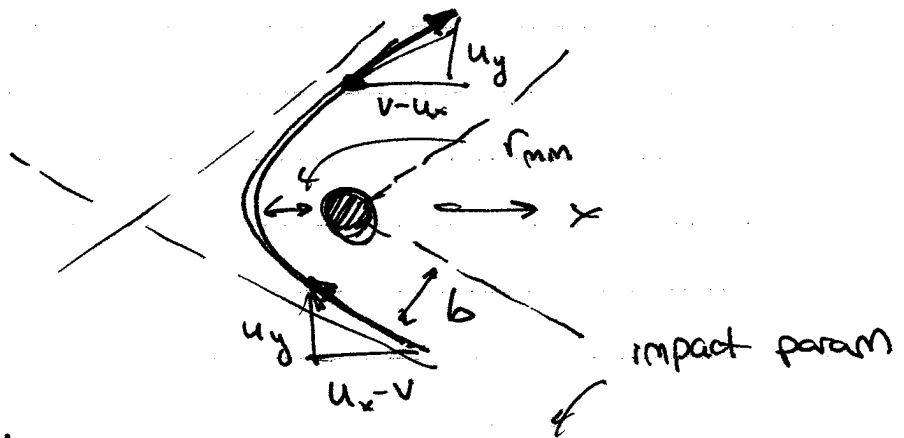
eg: Gravity assist: spacecraft boosted by a well-orchestrated sequence of "scatterings" from planets

consider: spacecraft in near collision course w/ planet  
 (eg: STAR CONEOL)



$v > u_x$   
 so planet is "catching up"

cm frame:  $\approx$  planet rest frame

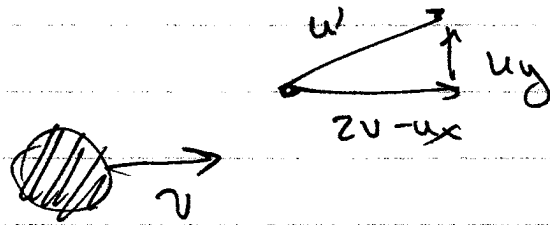


Mission control/cptn. preord: choose  $b$  s.t.

$x$ -axis of hyperbolic traj coincides w/ planet direction of motion



back to solar-system frame



compare final & initial spacecraft speeds:

$$\left(\frac{u'}{u}\right)^2 = \frac{(2v - u_x)^2 + u_y^2}{u^2}$$

$$= \left(\frac{2v}{u}\right)^2 - 4 \frac{v}{u} \frac{u_x}{u} + 1$$

$\underbrace{\qquad\qquad\qquad}_{\cos \theta}$

$$= \left(\frac{2v}{u} - 1\right)^2 + 4 \frac{v}{u} (1 - \cos \theta)$$

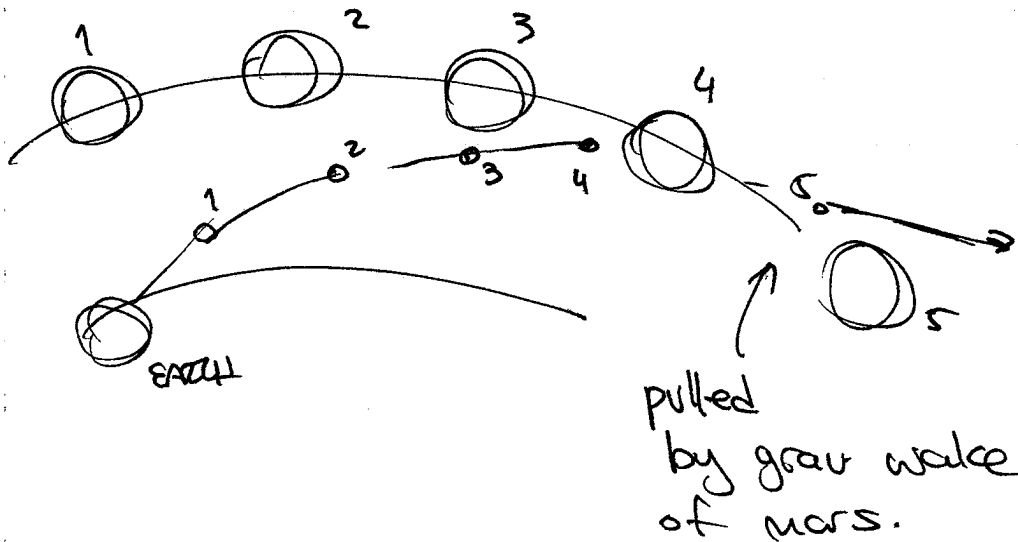
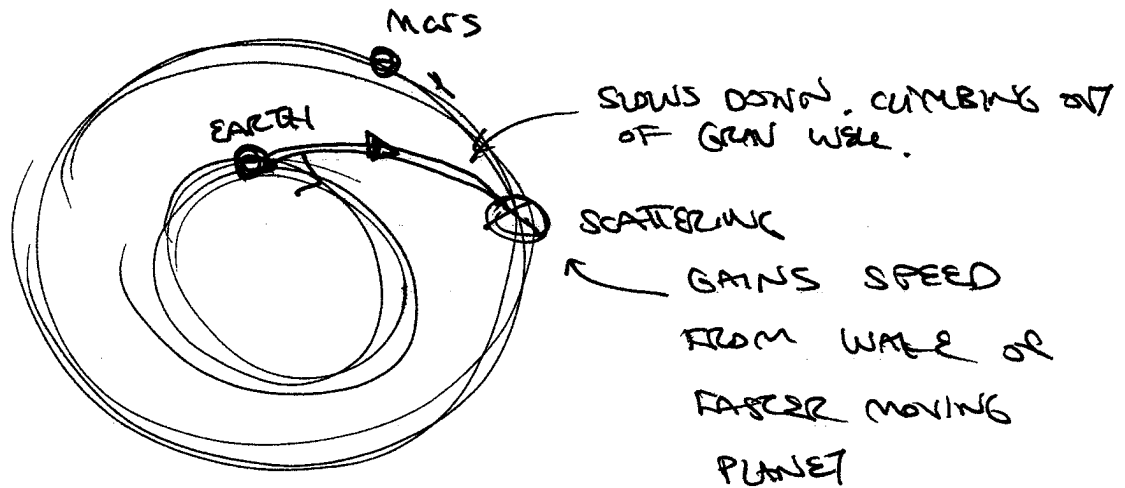
$u' > u$  for  $v' > u$

small angle  $\rightarrow$  greater boost

Q: how is it that energy is conserved?

# Voyager spacecraft

series of grav assists to climb out of sun's gravitational well



ORBIT ANALYZED @ 2 LENGTH SCALES

large: ELLIPSES AROUND SUN

small: hyperbolic, scattering off mars